



# AE/ME 339 Computational Fluid Dynamics (CFD)

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topic12b\_shock\_tube\_flow

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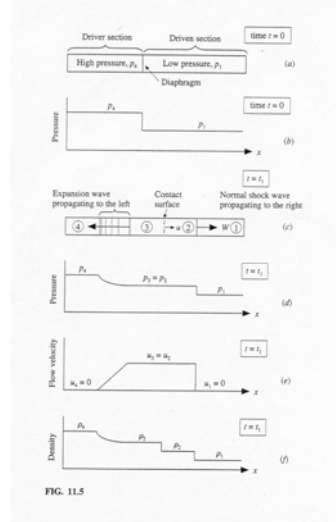
Computational Fluid Dynamics (AE/ME 339)

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Basic parameter of the shock tube is the diaphragm pressure ratio  $p_4/p_1$ .

The two chambers may be at different temperatures,  $T_1$  and  $T_4$ , and may contain different gases having different gas constants,  $R_1$  and  $R_4$ .

At the instant when the diaphragm is broken, the pressure distribution is a step function. It then splits into a shock and an expansion fan as shown in the figure.



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The shock propagates into the expansion chamber with speed  $V_{\text{shock}}$   
An expansion wave propagates into the high pressure chamber with speed  $a_4$  at its front.

Condition of the shock traversed by the shock is denoted by 2 and that traversed by the expansion wave is denoted by 3.

The interface between Regions 2 and 3 is called the *contact surface*. It marks the boundary between the fluids which were originally separated by the diaphragm. The contact surface is like the front of a piston driving into the low pressure region creating a shock front ahead of it.

The following conditions apply on either side of the contact surface

$$p_2 = p_3$$

and

$$u_2 = u_3$$

Temperatures and densities will be different in Regions 2 and 3. The above two conditions are used to determine the shock strength  $p_3/p_4$  and expansion strength  $p_2/p_1$ .

$$u_2 = a_1 \left( \frac{p_2}{p_1} - 1 \right) \sqrt{\frac{2/\gamma_1}{(\gamma_1 + 1)p_2/p_1 + (\gamma_1 - 1)}}$$

$$u_3 = \frac{2a_4}{\gamma_4 - 1} \left[ 1 - \left( \frac{p_3}{p_4} \right)^{(\gamma_4 - 1)/2\gamma_4} \right]$$

Equating the above two expression and using  $p_3 = p_2$  gives

$$\frac{p_4}{p_1} = \frac{p_2}{p_1} \left[ 1 - \frac{(\gamma_4 - 1)(a_1/a_4)(p_2/p_1 - 1)}{\sqrt{2\gamma_1} \sqrt{2\gamma_1 + (\gamma_1 + 1)(p_2/p_1 - 1)}} \right]^{-2\gamma_4/\gamma_4 - 1}$$

The above expression gives shock strength  $p_2/p_1$  implicitly as a function of the diaphragm pressure ratio  $p_4/p_1$

The expansion strength can then be obtained as

$$\frac{p_3}{p_4} = \frac{p_3}{p_1} \frac{p_1}{p_4} = \frac{p_2/p_1}{p_4/p_1}$$

The temperature behind the shock is obtained from the Rankine-Hugoniot relations

$$\frac{T_2}{T_1} = \frac{1 + \frac{\gamma_1 - 1}{\gamma_1 + 1} \frac{p_2}{p_1}}{1 + \frac{\gamma_1 - 1}{\gamma_1 + 1} \frac{p_1}{p_2}}$$

$$\frac{\partial}{\partial t} \left\{ \begin{array}{l} \rho \\ \rho u \\ \rho \left( e + \frac{u^2}{2} \right) \end{array} \right\} + \frac{\partial}{\partial x} \left\{ \begin{array}{l} \rho u \\ \rho u^2 + p \\ \rho \left( e + \frac{u^2}{2} \right) u + pu \end{array} \right\} = 0 \dots (A)$$